U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION TYPE CERTIFICATE DATA SHEET E26NE	TCDS NUMBER E26NE REVISION: 13* DATE November 19, 2007 PRATT & WHITNEY CANADA MODELS: PT6A-66 PT6A-67B PT6A-68 PT6A-66A PT6A-67D PT6A-67AG PT6A-67 PT6A-67R PT6A-67T PT6A-67A PT6A-64 PT6A--66D PT6A-67AF PT6A-66B PT6A-67F PT6A-67P

Engines of models described herein conforming with this data sheet (which is part of Type Certificate Number E26NE) and other approved data on file with the Federal Aviation Administration, meet the minimum standards for use in certificated aircraft in accordance with pertinent aircraft data sheets and applicable portions of the Federal Aviation Regulations, provided they are installed, operated, and maintained as prescribed by the approved manufacturer's manuals and other approved instructions.

TYPE CERTIFICATE (TC) HOLDER: Pratt & Whitney Canada, Inc.
(formerly Pratt & Whitney Aircraft of Canada, Ltd)
1000 Marie Victorin
Longueuil, Quebec
Canada J4G 1A1

I. MODELS	**PT6A-66**	**PT6A-67**	**PT6A-67A**	**PT6A-67R**	**PT6A-67T**	**PT6A-64**	**PT6A-66B**
TYPE	A free turbine turbo-propeller propulsion engine incorporating a multi-stage compressor driven by a single-stage turbine and a two-stage free turbine driving the propeller shaft through planetary reduction gearing.						
RATINGS (see NOTE 4)							
Maximum continuous at sea level							
Equivalent shaft horsepower	905	1,272	1,272	1,294	1,294	747	1010
Shaft Horsepower	850	1,200	1,200	1,220	1,220	700	950
Thrust, pounds	138	181	181	184	184	119	150
Output, rpm (maximum)	2,000	1,700	1,700	1,700	1,700	2,000	- -
Gas gen. rpm (maximum)	39,000	39,000	39,000	39,000	39,000	39,000	- -
							.
Takeoff (5 min. at sea level)							
Equivalent shaft horsepower	905	1,272	1,272	1,509	1,509	747	1010
Shaft horsepower	850	1,200	1,200	1,424	1,424	700	950
Thrust, pounds	138	181	181	212	212	119	150
Output, rpm (maximum)	2,000	1,700	1,700	1,700	1,700	2,000	- -
Gas gen., rpm (maximum)	39,000	39,000	39,000	39,000	39,000	39,000	- -
Alternate takeoff (5 min. at sea level)							
Equivalent shaft horsepower	---	---	---	1,358	---	---	---
Shaft horsepower	---	---	---	1,281	---	---	---
Thrust, lbs.	---	---	---	192	---	---	---
Output, rpm (maximum)	---	---	---	1,700	---	---	---
Gas gen. rpm (maximum)	---	---	---	39,000	---	---	---
Maximum Reverse							
Shaft horsepower	800	900	- -	- -	- -	700	800
Output, rpm (maximum)	1,900	1,650	- -	- -	- -	1,900	- -

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LEGEND: "- -" INDICATES "SAME AS PRECEDING MODEL"
"---" NOT APPLICABLE NOTE: ALL PAGES ARE REFORMATTED. SIGNIFICANT CHANGES IF ANY, ARE BLACK-LINED IN THE LEFT MARGIN.

Models

	PT6A-67B	PT6A-67D	PT6A-67AG	PT6A-68	PT6A-66A	PT6A-66D	PT6A-67AF	PT6A-67F	PT6A-67P
Type	A free turbine turbo-propeller propulsion engine incorporating a multi-stage compressor driven by a single-stage turbine and a two-stage free turbine driving the propeller shaft through planetary reduction gearing.								
RATINGS (see NOTE 4)									
Maximum continuous at sea level									
Equivalent shaft horsepower	1,272	1,285	1,294	1,324	905	- -	1294	1796	1272
Shaft Horsepower	1,200	1,214	1,220	1,250	850	- -	1220	1700	1200
Thrust, pounds	181	178	184	185	138	137	184	241	181
Output, rpm (maximum)	1,700	1,700	1,700	2,000	- -	- -	1700	- -	- -
Gas gen. rpm (maximum)	39,000	39,000	39.000	39,000	- -	- -	- -	- -	- -
Takeoff (5 min. at sea level)									
Equivalent shaft horsepower	1,272	1,353	1,430	1,324	905	- -	1509	1796	1972
Shaft horsepower	1,200	1,279	1,350	1,250	850	- -	1424	1700	1200
Thrust, pounds	181	186	200	185	138	137	212	241	181
Output, rpm (maximum)	1,700	1,700	1,700	2,000	- -	- -	1700	- -	- -
Gas gen., rpm (maximum)	39,000	39,000	39,000	39,000	- -	- -	- -	- -	- -
Alternate takeoff (5 min. at sea level)	---	---	---	---	---	---	---	---	---
Equivalent shaft horsepower	---	---	---	---	---	---	---	---	---
Shaft horsepower	---	---	---	---	---	---	---	---	---
Thrust, lbs.	---	---	---	---	---	---	---	---	---
Output, rpm (maximum)	---	---	---	---	---	---	---	---	---
Gas gen. rpm (maximum)									
Maximum Reverse	900	900	900	---	800	- -	900	- -	- -
Shaft horsepower	1,650	1,650	1,650	---	1900	- -	1650	- -	- -
Output, rpm (maximum)									

I. MODELS (Continued)	PT6A-66	PT6A-67	PT6A-67A	PT6A-67R	PT6A-67T	PT6A-64	PT6A-67B	PT6A-66B
Limitations								
Maximum Continuous								
Maximum interturbine temp (ITT)	1,526°F	1,526°F	1,544°F	1,544°F	1,544°F	1,472°F	1,472°F	1544 °F
Maximum air inlet temp (AIT) for rated power	135°F	115°F	127°F	119°F	92°F	135°F	113°F	147 °F
Takeoff (5 mins)								
Maximum interturbine temp (ITT),	1,526°F	1,544°F	1,562°F	1,571°F	1,571°F	1,472°F	1,472°F	1562 °F
Maximum air inlet temp(AIT) for rated power (AIT)	135°F	124°F	127°F	99°F	94°F	135°F	125°F	147 °F
Alternate takeoff (10 min)								
Maximum interturbine temp (ITT)	---	---	---	1,517°F	---	---	---	---
Maximum air inlet temp (AIT)for rated power (AIT)	---	---	---	91°F	---	---	---	---
Starting (5 secs)								
Maximum interturbine temp (ITT)	1,832°F	- -	- -	- -	- -	- -	- -	- -

I. MODELS (Continued)	PT6A-67B	PT6A-67D	PT6A-67AG	PT6A-68	PT6A-66A	PT6A-66D	PT6A-67AF	PT6A-67F	PT6A-67P
Limitations									
Maximum Continuous									
Maximum interturbine temp (ITT)	1,472°F	1436 F	1,472°F	1508°F	1472°F	1544°F	- -	1598 F	1544
Maximum air inlet temp (AIT) for rated power	113°F	115 F	92°F	108°F	122°F	158°F	119 °F	90 F	122
Takeoff (5 mins)									
Maximum interturbine temp (ITT),	1,472°F	- -	- -	1508°F	1472°F	1562°F	1571 °F	1598 F	1562
Maximum air inlet temp (AIT) for rated power	125°F	118 F	79°F	108°F	122°F	158°F	99 °F	90 F	111 F
Alternate takeoff (10 min)									
Maximum interturbine temp (ITT)	---	---	---		---	---	---	---	---
Maximum air inlet temp (AIT) for rated power	---	---	---		---	---	---	---	---
Starting (5 secs)									
Maximum interturbine temp (ITT)	1,832°F	- -	- -	- -	- -	- -	- -	- -	- -

Fuel Type

Fuels conforming to the current P&WC specifications CPW 204 and CPW 46. Refer to the current revision of Service bulletin 14004 for all models except for the PT6A-67AG and PT6A-68. For the PT6A-67AG and PT6A-67F refer to Service Bulletin 14504 and for the PT6A-68, refer to Service Bulletin 18004. Refer to the Installation Manuals for further details.

Oil Type

Oils conforming to PWA specification PWA 521 Type II. Refer to the current revision of P&WC Service Bulletin 14001 for acceptable lubrication for all models except the PT6A-68. For the PT6A-68, refer to Service Bulletin 18001.

Equipment

Fuel pump, fuel control unit, ignition system without power source, propeller governor and fuel heater are included as standard equipment as shown in the approved Parts List. For additional information refer to Installation Manual. For output drive specifications and C.G. location, refer to Installation Manual.

PT6A-	66	67	67A	67R	67T	64	66B	67B
Dimensions	**PRINCIPAL DIMENSIONS AT ROOM TEMPERATURE / INCHES / NOMINAL**							
Diameter	18.3	- -	- -	- -	- -	- -	- -	- -
Length	69.7	74.2	- -	75.2	75.2	69.7	- -	76.0
Weight	**WEIGHT / DRY / POUNDS / INCLUDING EXTERNAL ENGINE ACCESSORIES**							
Standard rotation	456	506	- -	515	- -	456	- -	517
Reverse rotation	470	---	---	---	---	---	470	
Reduction Ratio (Np:Nf)	0.0663:1	0.0568:1	- -	- -	- -	0.0663:1	- -	.0568:1

PT6A-	67D	67AG	68	66A	66D	67AF	67F	67P
Dimensions	**PRINCIPAL DIMENSIONS AT ROOM TEMPERATURE/ INCHES/ NOMINAL**							
Diameter	- -	- -	22.2	18.3	- -	- -	- -	-.-
Length	74.2	75.2	71.3	69.9	- -	75.2	- -	76.0
Weight	**WEIGHT/ DRY/ POUNDS/ INCLUDING EXTERNAL ENGINE ACCESSORIES**							
Standard rotation	515	520	572	456	- -	515	590	554
Reverse rotation								
Reduction Ratio (Np:Nf)	0.0568:1	- -	0.0667:1	0.0663:1	- -	0.0568:1	0.0575:1	0.0568:1

CERTIFICATION BASIS

All models except the PT6A-68 and PT6A-66D & PT6A-66B - FAR Part 33 effective February 1, 1965, and Amendments 33-1 to 33-10, inclusive. PT6A-68 – FAR Part 33 effective February 1, 1965, and Amendments 33-1 to 33-14 inclusive, including FAR Part 34 for turbo-propeller engines rated below 1000 kilowatts (1340 shp). PT6A-66D, PT6A-66B, PT6A-67F and PT6A-67P – FAR Part 33 effective February 1, 1965 and Amendments 33-1 to 33-20 inclusive.

Type Certificate E26NE	MODEL	APPLICATION DATE	ISSUED/REVISED
	PT6A-66	MAR 27, 1985	MAR 16, 1987
	PT6A-67	MAR 27, 1985	MAR 16, 1987
	PT6A-67R	AUG 05, 1985	MAR 16, 1987
	PT6A-67A	JUL 05, 1987	DEC 22, 1987
	PT6A-64	DEC 06, 1989	APR 09, 1990
	PT6A-67B	SEP 12, 1990	MAR 05, 1991
	PT6A-67D	SEP 12, 1990	MAR 05, 1991
	PT6A-67AG	MAY 16, 1994	JUNE 23, 1994
	PT6A-68	JUNE 11, 1996	JUNE 30, 1997
	PT6A-66A	NOV 10, 1998	MAR 11, 1999
	PT6A-67T	NOV 5, 1999	NOV 29, 2000
	PT6A-66D	MAY 17, 2005	NOV 21, 2005
	PT6A-66B	APRIL 3, 2006	JAN 29, 2007
	PT6A-67AF	MAY 17, 2006	JAN 29, 2007
	PT6A-67F	OCT 5, 2006	AUG 2, 2007
	PT6A-67P	FEB 9, 2007	NOV 19, 2007

IMPORT REQUIREMENTS

To be considered eligible for installation on U.S. registered aircraft, each engine (or propeller) to be exported to the United States shall be accompanied by a certificate of airworthiness for export or certifying statement endorsed by the exporting cognizant civil airworthiness authority and containing the following language.

(1) This engine (or propeller) conforms to its United States type design (Type Certificate Number E26NE) and is in a condition for safe operation.
(2) This engine (or propeller) has been subjected by the manufacturer to a final operational check and is in a proper state of airworthiness.

Reference FAR Section 21.500 which provides for the airworthiness acceptance of aircraft engines or propellers manufactured outside the U.S. for which a U.S. type certificate has been issued.

Additional guidance is contained in FAA Advisory Circular 21-23, Airworthiness Certification of Civil Aircraft, Engines, Propellers and Related Products, Imported into the United States.

NOTES									
I. MODELS	**PT6A-66**	**PT6A-67**	**PT6A-67A**	**PT6A-67R**	**PT6A-67T**	**PT6A-64**	**PT6A-67B**	**PT6A-67D**	**PT6A-66B**
NOTE 1	**MAXIMUM PERMISSIBLE ENGINE OPERATING SPEEDS (RPM)**								
Gas generator rotor speed									
Takeoff	39,000	- -	- -	- -	- -	- -	- -	- -	- -
Alternate takeoff	---	---	---	39,000	---	---	---	---	---
Maximum continuous	39,000	- -	- -	- -	- -	- -	- -	- -	- -
Transient	39,000	- -	- -	- -	- -	- -	- -	- -	- -
Power turbine rotor speed									
Takeoff	30,145	29,894	- -	- -	29,894	30,145	29,894	- -	30,145
Alternate takeoff	---	---	---	29,894	---	---	---	---	---
Maximum continuous	30,145	29,894	- -	- -	29,894	30,145	29,894	32,894	30,145
Transient	33,235	32,883	- -	- -	- -	33,235	32,883	- -	33,235

I. MODELS	**PT6A-66**	**PT6A-67**	**PT6A-67A**	**PT6A-67R**	**PT6A-67T**	**PT6A-64**	**PT6A-67B**	**PT6A-67D**	**PT6A-66B**
NOTE 2	**MAXIMUM PERMISSIBLE TEMPERATURES / DEGREES FAHRENHEIT**								
Interturbine temperature (ITT)									
Takeoff	1,526	1,544	1,562	1,571	1,571	1,472	1,472	1,472	1562
Alternate takeoff	---	---	---	1,517	---	---	---	---	---
Maximum continuous	1,526	1,526	1,544	1,544	1,544	1,472	1,472	1,436	1544
Starting (5 seconds)	1,832	- -	- -	- -	- -	- -	- -	- -	- -
Air inlet temperature(AIT)									
Takeoff									
Alternate takeoff	135	124	127	99	94	135	125	118	147
Maximum continuous	---	---	---	91	---	---	---	---	---
	135	115	127	119	92	135	113	116	147

NOTE 3	**MAXIMUM PERMISSIBLE OUTPUT TORQUE, FOOT-POUNDS**								
Takeoff	2,230	3,708	- -	4,400	- -	2,230	3,708	3,950	2,495
Alternate takeoff	---	---	---	3,960	---	---	---	---	---
Maximum continuos	2,230	3,708	- -	3,825	3,825	2,230	3,708	3,750	2,495
Transient (20 seconds)	2,750	5,100	- -	5,100	5,100	2,750	5,100	- -	2,750

I. MODELS (CONTINUED)	PT6A-67AG	PT6A-68	PT6A-66A	PT6A-66D	PT6A-67AF	PT6A-67F	PT6A-67P
NOTE 1	**MAXIMUM PERMISSIBLE ENGINE OPERATING SPEEDS (RPM)**						
Gas generator rotor speed							
Takeoff	39,000	- -	- -	- -	- -	- -	- -
Alternate takeoff	---	---	---	---	---	---	---
Maximum continuous	39,000	- -	- -	- -	- -	- -	- -
Transient	39,000	- -	---	- -	- -	- -	- -
Power turbine rotor speed							
Takeoff	29,894	29,981	30,145	- -	29,894	29,564	29,894
Alternate takeoff	---	---	---	---	---	---	---
Maximum continuous	29,894	29,981	30,145	- -	29,894	29,564	29,894
Transient	32,883	32,979	33,235	- -	32,883	32,520	32,883
NOTE 2	**MAXIMUM PERMISSIBLE TEMPERATURES / DEGREES FAHRENHEIT**						
Interturbine temperature (ITT)							
Takeoff	1472	1508	1472	1562	1571	1598	1562
Alternate takeoff	---	---	---	---	---	---	---
Maximum continuous	1472	1508	1472	1544	- -	1598	1544
Starting (5 seconds)	1832	- -	- -	- -	- -	- -	- -
Air inlet temperature (AIT)							
Takeoff	79	108	122	158	99	90	111
Alternate takeoff	---	---	---	---	---	---	---
Maximum continuous	92	108	122	158	119	90	122

I. MODELS (CONTINUED)	PT6A-67AG	PT6A-68	PT6A-66A	PT6A-66D	PT6A-67AF	PT6A-67F	PT6A-67P
NOTE 3	**MAXIMUM PERMISSIBLE OUTPUT TORQUE, FOOT-POUNDS**						
Takeoff	4,170	3,283	2,230	- -	4,400	5,252	3,708
Alternate takeoff	---	---	---	---	---	---	---
Maximum continuous	4,170	3,283	2,230	- -	4,400	5,252	3,708
Transient (20 seconds)	5,100	3,800	2,750	3,750	5,100	6,092	5,100

NOTE 4.

The engine ratings are based on dry sea level static ICAO standard atmospheric conditions. No external accessory loads and no air bleed. The quoted ratings are obtained on a test stand with the specified fuel and oil, without intake ducting and utilizing the exhaust port and intake defined in the approved installation manual.

NOTE 5.

Fuel pressure and temperature

FUEL AND OIL PRESSURE AND TEMPERATURE LIMITS

Fuel pressure and fuel temperature limitations are shown in the engine installation manual.

NOTE 5 (cont'd)	**OIL PRESSURE LIMITS**							
PT6A-	66	67	67A	67R	67T	64/ 66A	67B	67D
Takeoff (psig)	90-135	- -	- -	- -	- -	100-135	90-135	- -
Maximum continuous (psig)	90-135	- -	- -	- -	- -	100-135	90-135	- -
Transient (psig)	40-200	- -	- -	- -	- -	- -	--	- -
Minimum inflight (psig)	60	- -	- -	- -	- -	- -	- -	- -

PT6A-	67AG	68 (2)	66D	66B	67AF	67F	67P
Takeoff	90-135	90-120	100-135	90-135	- -	- -	- -
Maximum continuous	90-135	90-120	100-135	90-135	- -	- -	- -
Transient	- -	40-200	- -	- -	---	---	---
Minimum	60	40	60	- -	- -	- -	- -

(1) Gas generator speed 27,000 RPM or above and oil temperature (140-160°F)
(2) For the PT6A-68 during aerobatic flight, the allowable steady state oil pressure range is 40-130 psig. At idle power, the minimum allowable transient oil pressure is 15 psig for a maximum of 5 seconds.

Oil Temperature

PT6A-	66A	66/ 67	67A	67R	67T	64	67B	67D
	OIL TEMPERATURE LIMITS / DEGREES FAHRENHEIT							
Takeoff	32-219	32-230	- -	- -	--	32-219	32-230	--
Transient (1)	32-230	- -	- -	- -	--	- -	--	--
Maximum continuous	32-219	32-230	- -	- -	--	32-219	32-230	--
Minimum	(-40)	- -	- -	- -	- -	- -	- -	- -
	(1) For the PT6A-64, -66A and –66D, the transient time is limited to 10 minutes.							

PT6A-	67AG	68 (2)	66D	66B	67AF	67F	67P
	OIL TEMPERATURE LIMITS / DEGREES FAHRENHEIT						
Takeoff	- -	50-230	32-219	32-230	- -	50-230	32-230
Transient (1)	- -	32-230	32-230	- -	- -	- -	- -
Maximum continuous	- -	50-230	32-219	32-230	- -	50-230	32-230
Minimum	- -	- -	- -	- -	- -	- -	- -
	(1) For the PT6A-64, -66A and –66D, the transient time is limited to 10 minutes						

NOTE 6.	**ACCESSORY DRIVE PROVISIONS ALL MODELS**					
				TORQUE	MOMENT	TORQUE
		SPEED RATIO		CONTINUOUS	OVERHANG	STATIC
DRIVE	ROTATION (1)	POWER TURBINE	GAS GENERATOR	(lb-in)	(lb-in)	(lb-in)
Tachometer, accessory gearbox**	CCW		0.112/ .140***	7/ 11.5***	10	100
Starter/generator	CW		0.293	170	250	1600
Vacuum pump	CCW		0.102	60	25	800
Hydraulic pump	CCW		0.204	150	25	800
Aircraft Accessory Drive	CW		0.321	135/ 160***	25	800
Tachometer, reduction gearbox**	CW	.1264*/.1405		7	10	100
Propeller overspeed governor**	CW	.1264*/.1405		50	25	850
Propeller governor**	CW	.1264*/.1405		50	25	850

NOTE 6 (cont)	(1) Direction of shaft rotation, facing engine pad: CCW = Counterclockwise CW = Clockwise Gas generator speed (Ng) 100% = 37,468 RPM Power turbine speed (Np): PT6A-64/66/66A/66D/66B Np 100% = 33,235 rpm (propeller shaft speed 2,205 rpm) PT6A-67/67A/67B/67D/67R/67AG/67T/67AF/ 67P Np 100% = 29,894 rpm (propeller shaft speed ,1700 rpm) PT6A-68 Np100% = 29,906 rpm (propeller shaft speed 1,995 rpm) PT6A-67F Np 100% = 29,564 RPM (Propeller shaft speed 1700 RPM) * For the PT6A-64, -66, –66A, 66D & 66B series only ** Not applicable to the PT6A-68 *** For the PT6A-67F

NOTE 7. Applicable Maintenance Manuals (Pratt & Whitney Canada part numbers) are: Models PT6A-66/66A/66B, P/N 3036122; Models PT6A-67/67A/67R/67AG/67T/67AF, P/N 3036132; Model PT6A-64, P/N 3038321; Model PT6A-68, P/N 3040872; Model PT6A-66D, P/N 3070902; Model PT6A-67F, P/N 3071152, Model PT6A-67D, -67B, -67P, P/N 3038336. Until the applicable maintenance manual is available, engines shall be maintained in accordance with Pratt & Whitney Canada Preliminary Maintenance Instructions.

NOTE 8. Applicable Overhaul Manuals (Pratt & Whitney Canada part numbers) are: Models PT6A-66/66A/66B P/N 3036123; Models PT6A-67/67A/67R/67AG/67T/67AF P/N 3036133; Model PT6A-64, P/N 3038322; Models PT6A-67B/ 67D/ 67P, P/N 3038337, Model PT6A-68 P/N 3040873; Model PT6A-66D, P/N 3070903; Model PT6A-67F, P/N 3071153, Until the applicable Overhaul Manual is available, all overhauls must be performed by Pratt & Whitney Canada in accordance with "new engine" standards.

NOTE 9. The PT6A-66/66B engines may be overhauled or maintained as three modules. The gas generator module and the power section module are separated at the "C" flange. The nose section module (2nd-stage reduction gearbox) is separated at the "A" flange:

		PT6A-66B
Gas generator module (PT6A-66):	P/N 3036400	3072313
Power section module (PT6A-66):	P/N 3036900	3072314
Nose section module, standard rotation (PT6A-66):	P/N 3107180	3112380
Nose section module, reverse rotation (PT6A-66):	P/N 3108280	3112480

NOTE 10. The PT6A-67/67A/67B/67D/67R/64/67AG/67T/68/66A/66D, 67AF and PT6A-67F engines may be overhauled or maintained as two modules, the gas generator module and the power section module. The separation point is the "C" flange:

Gas generator module PT6A-67/67A/67R/64/67AG/66A/67T/67AF	P/N 3036400
Gas generator module PT6A-67B	P/N 3042300
Gas generator module PT6A-67D	P/N 3044900
Gas generator module PT6A-68	P/N 3047000
Gas generator module PT6A-66D	P/N 3071015
Gas Generator module PT6A-67F	P/N 3071197
Gas Generator module PT6A-67P	P/N 3072959
Power section module PT6A-67/67A	P/N 3036600
Power section module PT6A-67R/67AG/67T/67AF	P/N 3039300
Power section module PT6A-67B	P/N 3042500
Power section module PT6A-67D	P/N 3044700
Power section module PT6A-64/66A	P/N 3045300
Power section module PT6A-68	P/N 3047200
Power section module PT6A-66D	P/N 3071018
Power section module PT6A-67F	P/N 3071193
Power section module PT6A-67P	P/N 3075956

NOTE 11. These engines meet the requirements of FAR 33.68 for operation in icing conditions as defined in FAR 25 Appendix C when the intake system conforms with the Pratt & Whitney Canada Installation Manual instructions for inertial separation of snow and icing particles. The engines also meet the requirements of FAR 33.27 and do not require external armoring.

NOTE 12. Life limits for critical rotating components are published in Pratt & Whitney Canada Service Bulletin Number 14002, 14502 (-67AG and -67F), 18002 (-68 only), 14302 (-67AF only).

NOTE 13. The recommended engine operating time between overhauls is published in Pratt & Whitney Canada Service Bulletin Number 14003 (-67R/67D), 14503 (-67AG only), 14603 (-64/ -66/-67/-67A/-67B/-67E/-66A/-67T/- 66D/- 66B/ -67P) and 14303 (-67AF and -67F).

NOTE 14.

	BLEED AIR (Also refer to Installation Manual)													
PT6A-	**66/66A 66B**	**67**	**67A**	**67R**	**67T**	**64**	**67B**	**67D**	**67AG**	**68**	**66D**	**67AF**	**67F**	**67P**
Maximum external (%)	7.5	5.25	- -	- -	- -	7.5	8.0	5.25	- -	- -	7.5	5.25	- -	8.0
Maximum during start (lb/min)	1.5	- -	- -	- -	- -	- -	- -	- -	- -	- -	- -	- -	- -	- -

NOTE 15. The PT6A-67R model includes provisions for automatic power increases from Alternate Takeoff Power to Takeoff Power.

NOTE 16. All models meet fuel venting requirements of SFAR 27, effective February 1, 1974, as amended by Amendments SFAR 27-1 through SFAR 27-4.

NOTE 17. Oil tank usable volume:
1.5 U.S. gallons / 5.68 litres / 1.25 imperial gallons.

For PT6A-68,
0.79 U.S. gallons / 3.00 litres / 0.66 imperial gallons, normal operation,
0.26 U.S. gallons / 1.00 litre / 0.22 imperial gallons, aerobatic manoeuvres.

Oil tank total capacity:
2.5 U.S. gallons / 9.46 litres / 2.08 imperial gallons.
4.23 U.S. gallons / 16.00 litres / 3.52 imperial gallons for PT6A-68.

NOTE 18. PT6A-67AG is a special purpose version of the PT6A-67 series of engines intended for use in agricultural aviation. This model may not be re-designated for other than agricultural operations. Engines intended for use in military and firefighting aviation, this model may not be re-designated for other then military or firefighting operations.

NOTE 19. Service Bulletins, structural repair manuals, vendor manuals, aircraft flight manuals, and overhaul and maintenance manuals, which contain a statement that the document is Transport Canada approved, are accepted by the FAA and are considered FAA approved. These approvals pertain to the type design only.

NOTE 20. The above models incorporate the following characteristics:

Model	Characteristics
PT6A-67	Basic turboprop model
PT6A-67A	A version of the basic -67 model with increased normal and maximum cruise ratings.
PT6A-66	Variant model limited to 850 SHP with 2000rpm standard and opposite rotation gearboxes.
PT6A-67R	Derivative of PT6A-67, 1424 SHP with 1700 rpm standard rotating gearbox and inclined turbine exhaust case.
PT6A-64	Variant model. Derivative of the PT6A-66 with the PT6A-61 reduction gearbox. Limited to 700 SHP with 2000 rpm standard rotation gearbox.
PT6A-67B	Variant model. Derivative of the basic -67 model using an upgraded reduction gearbox, flat rated at 1200 SHP.
PT6A-67D	Variant model. Derivative of the basic -67 model, similar to the PT6A-67R but with take-off flat rated to 1279 SHP.
PT6A-67AG	Variant model. Derivative of the basic -67 model, similar to other PT6A-67R but limited to 1350 SHP for special applications.
PT6A-66A	Derivative of the PT6A-66 with the PT6A-61 reduction gearbox. Limited to 850 SHP with 2000rpm, standard rotation gearbox.
PT6A-67T	Derivative of the basic -67 model, similar to the PT6A-67R but with the -67D hot end hardware.
PT6A-68	Derivative of the PT6A-67 family with a 2000 rpm, 1250 SHP gearbox and an electronic Power Management System with mechanical back-up. The engine also features a lubricating system capable of aerobatic maneuvers.
PT6A-67AF	Derivative of the basic A67 model using the A67R 1424 SHP take off rating for special applications.
PT6A-66B	Derivative of the PT6A-66 with the PT6A-67A thermal rating. Limited to 950 SHP with 2000 RPM standard & opposite rotation gearboxes.
PT6A-66D	Derivative of the PT6A-66A, with the PT6A-67A thermal rating. Limited to 850 SHP with 2000 RPM, standard rotation gearbox.
PT6A-67F	Derivative of the PT6A-67B and PT6A-68C with a new Reduction Gearbox Rated to 1700 SHP and a new Accessory Gearbox.
PT6A-67P	Derivative of the Pt6A-67B with the -67A thermal rating. The AGB has a mounting provision for a second generator unit.

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